B.) AMENDMENTS TO THE SPECIFICATION

Please replace Paragraphs [0014], [0015] and [0027] with the following new Paragraphs 14, 15 and 27:

[0014] The present invention is also a method of manufacturing a ceramic matrix composite turbine component. The method of the present invention includes providing at least one core insert section having a preselected geometry, the at least one core insert section comprising a material selected from the group consisting of silicon carbide-silicon carbide composite preform having at least some porosity, silicon-silicon carbide composite, silicon-silicon carbide composite preform having at least some porosity, ~~silicon-silicon carbide composite,~~ and a monolithic ceramic. The process also includes providing a plurality of plies of silicon carbide prepreg cloth and laying up a preselected number of silicon carbide prepreg plies to form at least one outer shell section. At least one at least one core insert section is assembled with the at least one outer shell section into a turbine blade form, wherein the turbine blade form comprises a dovetail section and an airfoil section, and wherein the at least one core insert section is positioned in the dovetail section of the turbine blade form. The turbine blade form is then autoclaved. Remaining porosity in the turbine blade form is filled with at least silicon using a silicon melt infiltration process. The filler material also forms a bond between the at least one core insert section and the at least one outer shell preform.

[0015] The present is also another method of manufacturing a ceramic matrix composite turbine blade component. The method of the present invention includes providing a core insert section having a preselected geometry, the core insert section comprising a material selected from the group consisting of silicon carbide-silicon carbide composite preform having at least some porosity, silicon-silicon carbide composite, silicon-silicon carbide composite preform having at least some porosity, ~~silicon-silicon carbide composite,~~ and a monolithic ceramic. An outer shell section preform having at least some porosity is provided. The at least one core insert section and the at least one outer shell preform are laid up into a turbine blade form, the turbine blade form comprising a dovetail section and an airfoil section, wherein the at least one core insert section is positioned in the dovetail section of the turbine blade form. Remaining porosity in the turbine blade form is filled with at least silicon using a silicon melt infiltration process. The filler material also forms a bond between the core insert section and the outer shell preform.

[0027] FIG. 1 depicts an exemplary aircraft engine LPT blade 20. In this illustration a turbine blade 20 comprises a ceramic matrix composite material. The turbine blade 20 includes an airfoil 22 against which the flow of hot exhaust gas is directed. The turbine blade 20 is mounted to a turbine disk (not shown) by a dovetail 24 that extends downwardly from the airfoil 22 and engages a slot of similar geometry on the turbine disk. The LPT blade of the present ~~invnetion~~ invention does not have an integral platform. A separate platform is provided to minimize the exposure of the dovetail to hot gases of combustion. The airfoil may be described as having a root end 40 and an oppositely disposed tip end 32.